

P 23

NASA CASE NO. LAR 15297-1

PRINT FIG. 2

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Serial No.: 08/492,686

Filed: 06/20/95

(NASA-Case-LAR-15297-1)  
SIMULTANEOUS LUMINESCENCE PRESSURE  
AND TEMPERATURE MAPPING SYSTEM  
Patent Application (NASA. Langley  
Research Center) 23 p

LaRC

N96-12093

Unclass

63/70 0071242

PAT. APPL.  
IN-70  
AUMC. CASE

### SIMULTANEOUS LUMINESCENCE PRESSURE AND TEMPERATURE MAPPING SYSTEM

The present invention relates generally to pressure and temperature mapping systems, and more particularly to simultaneous temperature mapping systems for use at high temperatures.

A simultaneous luminescence pressure and temperature mapping system is developed including improved dye application techniques for surface temperature and pressure measurements from 5 torr to 1000 torr with possible upgrade to from 0.5 torr to several atmospheres with improved camera resolution. Adsorbed perylene dye on slip-cast silica is pressure (oxygen) sensitive and reusable to relatively high temperatures (~150°C). Adsorbed luminescence has an approximately linear color shift with temperature, which can be used for independent temperature mapping and brightness pressure calibration with temperature.

Novel aspects of the present invention include enabling non-intrusive, simultaneous pressure and temperature measurements using a single data acquisition system. Additionally, this technology allows for testing at higher temperatures than is currently available.

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Serial No.: 08/492,686  
Filed: 06/20/95

SIMULTANEOUS LUMINESCENCE PRESSURE  
AND TEMPERATURE MAPPING SYSTEM

Origin of the Invention

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The invention described herein was made by an employee of the United States Government and may be used by and for the Government for governmental purposes without the payment of any royalties thereon or therefor.

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Background of the Invention

1. Technical Field of the Invention

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The present invention relates generally to pressure and temperature mapping systems, and more particularly to simultaneous temperature mapping systems for use at high temperatures.

2. Discussion of the Related Art

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Advanced diagnostic techniques capable of nonintrusive global measurement of aerodynamic pressure and heating distributions on models tested in hypersonic wind tunnels will significantly enhance the design and assessment process of future aerospace vehicles. Considerable progress  
25 has been made in measuring global temperature distributions with time, hence global heating distribution, via relative intensity thermographic phosphor and infrared emission techniques, but global pressure measurements at hypersonic conditions continue to elude experimental aerothermodynamicists. Pressure mapping techniques using  
30 photoluminescence offer a potential savings in time and expense for wind-

tunnel testing over conventional pressure instrumentation (i.e. discrete measurements via transducers) and are sought for hypersonic wind-tunnel applications.

5 Luminescence pressure mapping techniques have been developed and tested broadly by government agencies, industries, and universities for lower-speed applications. Current techniques use oxygen sensitive luminophores in polymer binders for pressure sensitive paints which can be applied to a variety of surfaces. Paint applications, however, are limited in high-temperature stability and time response by polymers. Temperature  
10 effects on induced luminescence further complicate pressure measurements in high-speed application with aerodynamic heating. Recent work was done to calibrate temperature effects on paint luminescence for wind-tunnel measurements using separate temperature mapping data. The results of this work and similar applications may expand near-term testing capabilities for  
15 relatively low enthalpy, long test duration facilities, but innovative solutions are needed for the bulk of hypersonic testing (i.e.  $M > 6$ , moderate to high enthalpy, blowdown and impulse facilities) including an internal method for thermal calibration, higher-temperature stability and faster response times.

It is accordingly an object of the present invention to provide a system  
20 for simultaneously mapping temperature and pressure.

It is another object of the present invention to provide a system for simultaneously mapping temperature and pressure at high temperatures.

It is a further object of the present invention to provide a system for simultaneously mapping temperature and pressure with reduced response  
25 time.

It is a further object of the present invention to provide a system for simultaneously mapping temperature and pressure which is suitable for use in hypersonic facilities.

It is yet another object of the present invention to accomplish the  
30 foregoing objects in a simple manner.

Additional objects and advantages of the present invention are apparent from the drawings and specification which follow.

#### Summary of the Invention

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According to the present invention, the foregoing and additional objects are obtained by providing a method for simultaneous mapping of temperature and pressure on an member, including the steps of applying a luminescent material, preferably perylene, to the surface of the member, exciting the luminescent material, measuring the resulting emissions, and  
10 determining the pressure and temperature on the surface of the member. The luminescent material may be excited using a blue light or long-wave ultraviolet light and the resulting emissions are measured using a two-color imaging system. The preferred method for applying the luminescent material  
15 is polishing the surface of the member, saturating the surface with the luminescent material, heating the saturated member, and repolishing the heated member. The member may be a slip-casted ceramic.

#### Brief Description of the Drawings

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Fig. 1 is a figure showing the mounting of test samples;

Fig. 2 is a graph showing emission brightness vs. pressure model at constant temperatures;

Fig. 3 is a graph showing ratio difference vs. temperature;

25

Fig. 4 is a graph showing color ratio data versus temperature;

Fig. 5a is a graph showing temperature sensitivity and quenching constants vs. adsorbed film thickness;

Fig. 5b is a graph showing reference color ratio vs. adsorbed film thickness;

30

Fig. 6a is a graph showing time vs. data points;

Fig. 6b is a graph showing irradiance vs. data points;

Fig. 6c is a graph showing pressure vs. data points;

Fig. 6d is a graph showing temperature vs. data points;

Fig. 7a is a graph showing color ratio vs. data points for sample region

5 2;

Fig. 7b is a graph showing brightness vs. data points for sample region

2;

Fig. 7c is a graph showing pressure vs. data points for sample region

2;

10 Fig. 7d is a graph showing temperature vs. data points for sample  
region 2;

Fig. 8a is a graph showing color ratio vs. data points for sample region

3;

Fig. 8b is a graph showing brightness vs. data points for sample region

15 3;

Fig. 8c is a graph showing pressure vs. data points for sample region

3;

Fig. 8d is a graph showing temperature vs. data points for sample  
region 2;

20 Fig. 9a is a graph showing reference brightness vs. data points;

Fig. 9b is a graph showing reference color ratio vs. data points;

Fig. 9c is a graph showing luminescence pressure vs. data points;

Fig. 9d is a graph showing temperature vs. data points;

25 Fig. 10 is a graph showing relative pressure uncertainty for various  
imaging resolutions; and

Fig. 10b is a graph showing absolute pressure uncertainty for various  
imaging resolutions.

Detailed Description of the Invention

The effects of oxygen interaction on luminescence yields and lifetimes are well documented for many substances since oxygen is an active  
5 quencher of fluorescence and phosphorescence. For some materials, quenching rates and efficiencies are large enough to make them useful as oxygen indicators. Theory predicts the quenching process to obey the Stern-Volmer equation:

10

$$\frac{I}{I_0} = \frac{1}{1 + K[O_2]} \quad (1)$$

15 for constant temperatures, where  $I$  and  $I_0$  are the fluorescence intensities in the presence and absence, respectively, of oxygen. In a mixed gas such as air, the oxygen concentration  $[O_2]$  is simply replaced by the partial gas pressure.  $K$  is the oxygen quenching coefficient which can be calibrated for many different luminescent materials.

20 Few guidelines exist for luminescence temperature dependence in oxygen sensors. Temperature is known, however, to affect quantum yield, quenching (Stern-Volmer) coefficient, and solubility and diffusion rates in binders.

Following initial luminescence measurements and calibration on  
25 ceramic test samples, perylene dye was applied to ceramic models and tested in a hypersonic wind tunnel. Models tested included a 0.02 scale model of a proposed personnel launch vehicle (HL-20) and a 0.0075 scale shuttle orbiter model. Both models represent configurations that had been previously slip-cast and tested using two-color luminescence thermography  
30 and infrared mapping techniques. The luminescent perylene dye

(Greengold) has reported green emission (520 nm) with blue-light excitation (460 nm) and is a commercially manufactured product used in coloring plastics. In adsorbed films on silica, an additional component of red emission was found on prepared samples.

5           Models were tested in a 31-inch Mach 10 wind tunnel. Measurements were made with a two-color imaging system which was developed for luminescence thermography. Simultaneous green and red filtered images of dye luminescence were recorded in which camera outputs are represented by

$$I_g = \int s_g(\lambda) I(\lambda) d\lambda \quad (2)$$

$$I_r = \int s_r(\lambda) I(\lambda) d\lambda \quad (3)$$

15 where  $I(\lambda)$  is the surface emission intensity with wavelength and  $s_g(\lambda)$  and  $s_r(\lambda)$  are the green and red detector sensitivities, respectively. Dye application and testing provided mixed results. Testing primarily demonstrated a need for improved model preparation and application

20 techniques. Perylene dye was applied to models in a solution with toluene via dipping, spray and brushing; however, in each case measurement anomalies from ceramic contaminants were apparent or quenching coefficients were significantly lowered. Later studies show a more preferred method for applying the dye which is described below.

25 Although quenching coefficients were minimal, adsorbed dye  
luminescence was shown to be stable except in stagnation regions which  
exceeded 200°C surface temperatures in three second run times (Mach 10  
flow, stagnation temperature > 730°C, stagnation pressure > 8600 KPa).  
Long-wave ultraviolet (UV)(365 nm) excitation was also used for irradiance  
30 over model surface areas with bright luminescence. Quantum efficiencies



are known to be lower for UV than for blue light excitation (~12%); however, advantages in using UV are readily accessible lamp supplies (mercury-arc lamps), which are in current use for luminescence thermography, less cost than with blue-light lasers or filtered Xenon lamps, and simpler and more efficient image filtering with better excitation-emission wavelength separation. Effects of photodegradation were also shown to be negligible.

Testing was performed in a vacuum oven chamber for development of dye application methods. Adsorbed films were previously applied to samples with a metered dropper which could not be replicated on test model surfaces. A subsequent technique was developed for consistent application. Additionally, effects of dye concentration were analyzed to determine optimum application thickness. Calibration measurements are shown for test samples and range and resolution of pressure measurements are determined.

For consistent dye application on ceramic matrix, a model is first polished with successively finer grits of silicon-carbide paper to a highly-smooth finish. Polishing residue is removed with solvent. The surface is then saturated by brush or dipped in a specific dye concentration with toluene. It is then allowed to dry and baked in a clean convection oven for thirty minutes at 160°C. Saturation and baking are repeated to ensure uniform application. Finally, the surface is re-polished with silicon-carbide paper removing accumulated dye layers on the exposed surface, hence, isolating film adsorption to ceramic pores which are more directly regulated by saturated dye concentration.

Ceramic disk samples were machined, 1-mm thick, from slip-cast silica rods 1.27-cm in diameter and drilled with 0.60-mm holes for mounting. Following dye application, samples were mounted as shown in Fig. 1 to electrical connectors in a ceramic block with butt-welded 0.076-mm diameter type K thermocouple wire and 0.51-mm diameter leads. As many as 5 test samples were mounted per test.

Through a test chamber window, image sampling regions are indicated in Fig. 3 with numbers 1 through 4. Region 1 is taken from an analysis sample prepared with low dye concentration. Regions 2 and 3 are from a test sample with optimum dye concentration and region 4 is taken from a test sample with much greater dye concentration. The test chamber was constructed from a 20.5-cm diameter pipe, lined with 2.54-cm thick low-density alumina ceramic insulation, fitted with an electrical resistance heater and o-ring sealed with a 2.54-cm thick quartz viewing window.

Chamber pressure was controlled via a vacuum pump and venting to atmosphere and was measured with a 1000 torr variable capacitance pressure transducer and Hastings vacuum gage down to 0.1 torr. Chamber insulation was effective in providing stable temperatures up to 150°C, taking approximately 4 to 6 hours for the chamber to cool from 150°C to room temperature at 0.5 torr.

Measurements made in the vacuum oven revealed an approximately linear shift in quantum efficiency  $I_0$  and quenching coefficient  $K$  in Equation (1) with temperature. Experimental data could be approximated to first order from Eqn. (1) for green brightness measurements as

$$\frac{I_g}{I_{00}} = \frac{(1+aT)}{1+K_0(1+bT)P} \quad (4)$$

where  $I_{00}$  is a calibration coefficient for reference brightness at zero pressure and temperature,  $a$  and  $b$  are temperature constants for quantum yield and oxygen quenching and  $K_0$  is the quenching constant at zero temperature with  $P$  being measured air pressure, assuming constant oxygen ratio. A plot of Eqn. (4) with experimentally derived constants is shown in Fig. 2 for air pressure at constant temperatures.

An approximately linear shift in emission color with temperature was observed from adsorbed perylene dye from 20°C to 150°C with a two-color imaging system. This characteristic was not expected but is highly desirable for temperature measurement and calibration. This allows for development of  
5 a system with a single dye application and simplifies filtering of broadband emission for diagnostics. In addition, color-ratio measurements were found to be independent of pressure and excitation irradiance. A linear approximation for green-to-red brightness ratio with temperature is given as

10

$$\frac{I_g}{I_r} - R_0 = cT \quad (5)$$

where c is the color-ratio temperature sensitivity and was found to be  
15 constant over a broad range of applications.  $R_0$  is a calibration coefficient for color ratio at zero temperature and was found to vary from sample to sample with apparent film thickness. A plot of Eqn. (5) is shown in Fig. 3 for temperature with experimentally derived constant c.

Image sampling regions consisted of 10 x 10 element pixel arrays and  
20 were averaged and treated as point data. Individual brightness response was calibrated separately for each sample region and color using a calibrated uniform intensity light source, or integrating sphere, with micrometer controlled aperture brightness and 2855K color temperature. (Photometric units in candela/m<sup>2</sup>).

25 Data was taken simultaneously for regions 1 through 3 in a recent experiment (see Fig. 6), and from a previous experiment for region 4, in which temperature, pressure and excitation irradiance were varied. Experiment results are shown in Fig. 4 for color ratio data versus temperature. Color-ratio bias (coefficient  $R_0$  in Eqn. (5)) is shown to vary from regions 2 to 3 on  
30 the same sample disk. Sensitivity coefficient (c in Eqn. (5)) is shown,

however, to be constant for regions 2 through 4, approaching zero in region 1 for thinner adsorption. Determination of quenching constants showed similar values for regions 1 through 3 ( $K_0=0.0011 \text{ torr}^{-1}$ ), and much lower value for region 4 ( $K_0=0.0005 \text{ torr}^{-1}$ ).

5           From this data and similar observations Figs. 5(a) and (b) were constructed as a guide for future dye applications. Stable values of temperature sensitivity and quenching constants ( $c$  and  $K_0$ ) are shown to exist over a range of film thicknesses corresponding to a range of calibration coefficient  $R_0$  roughly from 0.9 to 1.4.

10           Independent effects on quenching coefficient also include ceramic contaminants, interference with humidity and other gases, and later surface contaminants from handling and tunnel environments. Possible humidity effects are noted but are expected to be less of a concern in dry tunnel air supplies. Later surface contaminants may also be removed with most  
15           alcohols since the perylene dye is not easily soluble.

          Test conditions for recent experiments are shown in Figs. 6. (a) to (d). Data series were taken from four contiguous runs made over a three-day period. The longest run was several hours in duration. The first run began shortly after installing samples and evacuating the test chamber. From data  
20           points zero through 56, the vacuum oven was sealed except through a vent to atmosphere during pressure sweeps. Between points 56 and 59, the chamber was vented to atmosphere overnight with the quartz window removed. Points 59 to 61 were taken during re-evacuation from atmosphere to vacuum for the final run. A gap is shown between data points 61 and 64 to  
25           indicate changes made in camera gains.

          UV irradiance values shown in Fig. 6(b) were monitored using a photoelectric detector attached to the window perimeter. A substantial drift is shown in irradiance values over the longest run. This detector was used in observations only and not in subsequent calculations. Irradiance was

purposefully decreased for data points 12 through 14 using neutral density filters (N.D. = 0.1, 0.2, and 0.3) in the beam path.

Figs. 7 and 8 show the calibrated camera output in color ratio and individual brightness measurement for sample regions 2 and 3, respectively.

- 5 It is noted that color ratio measurements are independent of pressure and irradiance and seem to follow closely thermocouple measurements. Apparent drifts, however, are shown between the color ratio and thermocouple measurements at constant temperatures, which seem to be associated with initial pump down from atmosphere. This may be attributed
- 10 to atmospheric humidity, which was noticeably greater before the final run evacuation in comparison with initial run evacuation. These effects do not appear to be associated with cycling temperature or other forms of degradation. Suggested methods for wind-tunnel applications would include pulling down test section pressure several hours before run times to stabilize
- 15 luminescence parameters. Maximum drift rates as demonstrated, however, are well within wind-tunnel run rates.

In Fig. 9 luminescence pressures and temperatures are determined using periodic "wind-off" calibrations ( $I_{00}$  and  $R_0$ ) at reference data points separately for regions 2 and 3. Previously noted drifts are shown in both

20 reference brightness and color-ratio calibration coefficients calculated at periodic intervals. The corresponding pressure and temperature model used in calculations ( $K_0$ ,  $a$ ,  $b$  and  $c$ ) was shown in Figs. 1 and 2.

Sensible pressure range as well as resolution is primarily limited by the dynamic resolution of the imaging system. In present measurements,

25 video output is effectively digitized to less than 8-bits. Pressure measurement uncertainty can be derived as a quadratic function of pressure multiplied by relative brightness resolution from Eqn. (1), at constant temperature as

$$dP = \frac{dl}{I_0} \frac{(1+KP)^2}{K} \quad (6)$$

- 5 Values of  $I_0/dI$  used in uncertainty calculations were 200, 1000 and 4000, respectively, for the present imaging system and performance estimates for 10-bit and 12-bit systems. Figs. 10 (a) and (b) show calculations using Eqn. (6), and  $K=0.001 \text{ torr}^{-1}$  at  $25^\circ\text{C}$ , for relative pressure uncertainty in vacuum and absolute uncertainty over full range. Calculations show that the present  
10 system is operable from 5 torr to 1000 torr with greater than 1 signal-to-noise and from 0.5 torr to several atmospheres with higher imaging resolution.

Relatively bright luminescence was observed from adsorbed perylene dyes with blue-light excitation for temperatures up to  $150^\circ\text{C}$  and greater in vacuum and atmosphere, and was measured using standard video.

- 15 Comparative applications of a porphyrin dye were vaporized or permanently quenched under similar test conditions. A plot of Eqn. (4) with experimentally derived constants is shown in Fig. 2 for air pressure at constant temperatures.

- In subsequent testing and later in wind-tunnel applications, pressure  
20 and temperature data are derived from Equations (4) and (5) in the following manner. First it is assumed that  $K_0$ ,  $a$ ,  $b$ , and  $c$  are constants which can be applied to all points on a sample or model surface. In a typical applications, a "wind-off" measurement is first made for static pressure, temperature and separate color brightness

25

$$P_1, T_1, I_{g_1}(x, y), I_{r_1}(x, y) \\ \rightarrow I_{00}(x, y), R_0(x, y)$$

to determine reference brightness and color ratio coefficients at individual points on a surface. Then, during subsequent testing with flow conditions, color brightness images are recorded and processed with

$$I_{g_2}(x, y), I_{r_2}(x, y), [I_{00}(x, y), R_0(x, y)] \\ -P_2(x, y), T_2(x, y)$$

- 5 pre-run reference brightness and color ratio coefficients to determine in-situ pressures and temperatures on a surface.

Many improvements, modifications, and additions will be apparent to the skilled artisan without departing from the spirit and scope of the present invention as described herein and defined in the following claims.

- 10 What is claimed is:

SIMULTANEOUS LUMINESCENCE PRESSURE  
AND TEMPERATURE MAPPING SYSTEMAbstract of the Disclosure

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A simultaneous luminescence pressure and temperature mapping system is developed including improved dye application techniques for surface temperature and pressure measurements from 5 torr to 1000 torr with possible upgrade to from 0.5 torr to several atmospheres with improved camera resolution. Adsorbed perylene dye on slip-cast silica is pressure (oxygen) sensitive and reusable to relatively high temperatures (~150°C). Adsorbed luminescence has an approximately linear color shift with temperature, which can be used for independent temperature mapping and brightness pressure calibration with temperature.

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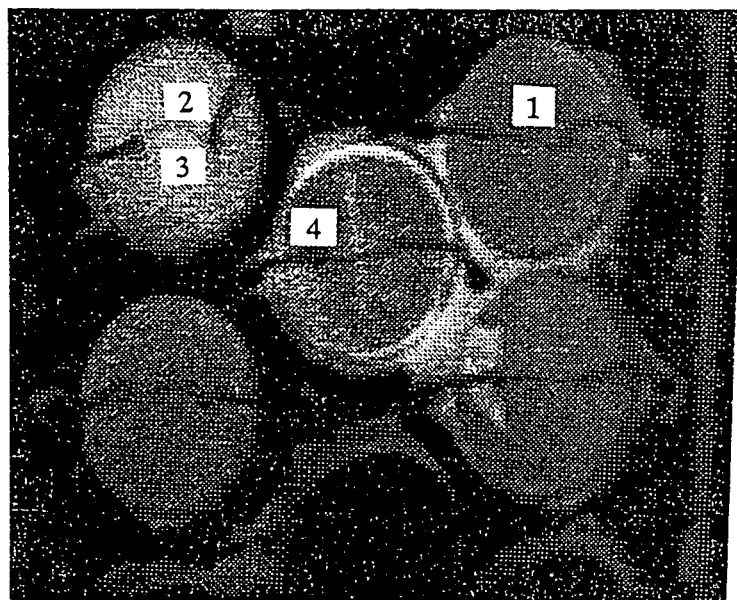


Fig. 1

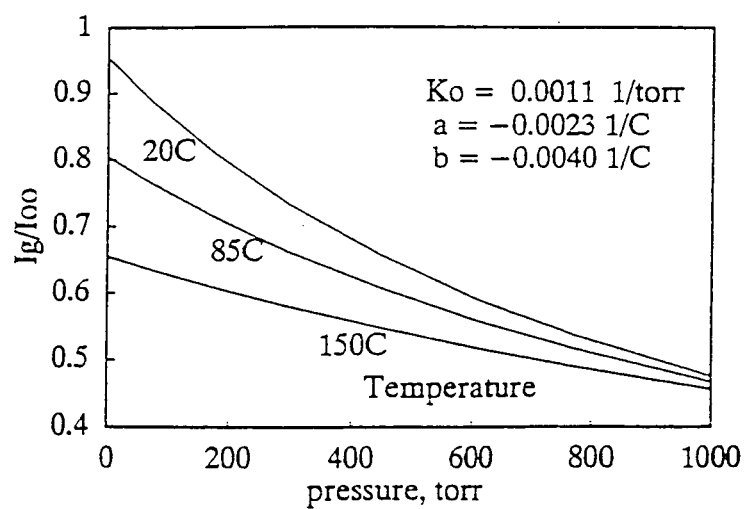


Fig. 2

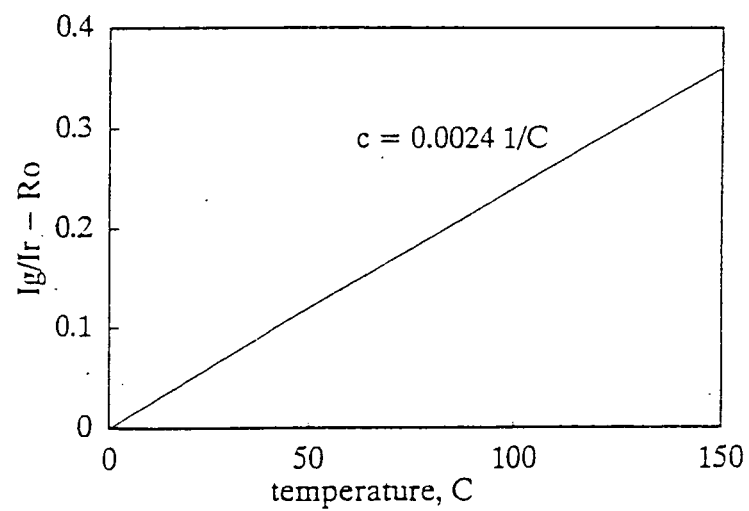


Fig. 3

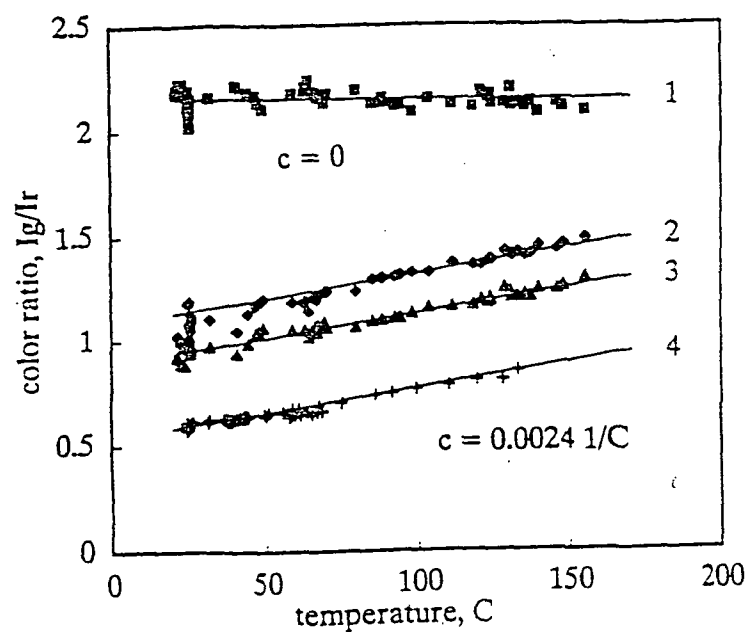


Fig. 4

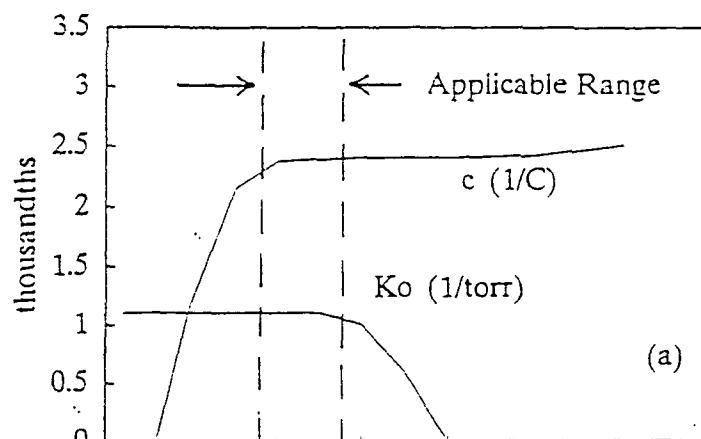


Fig. 5a

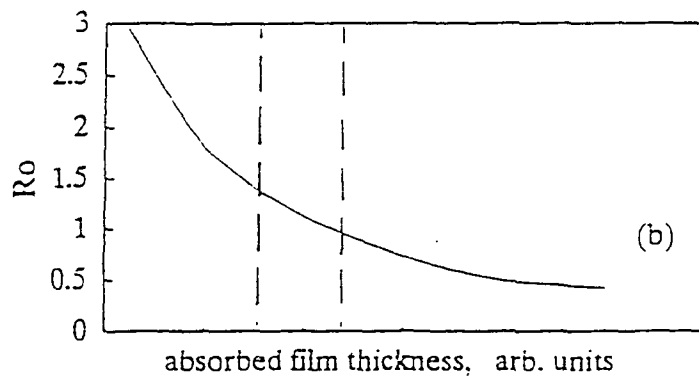


Fig. 5b

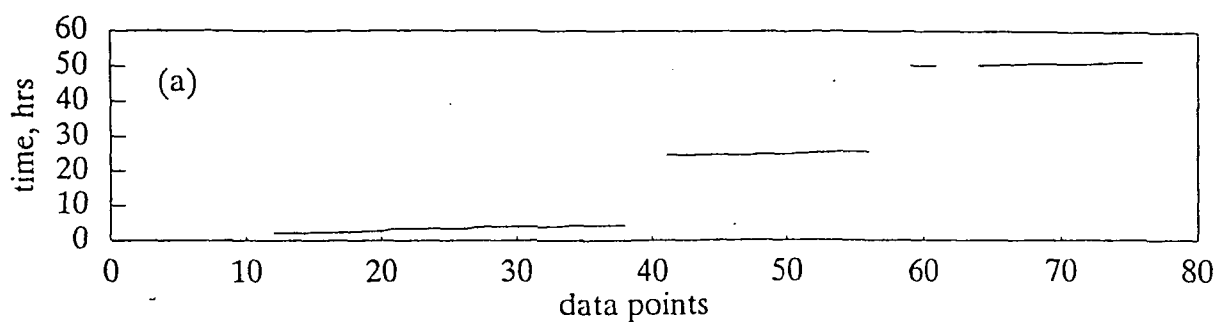


Fig. 6a

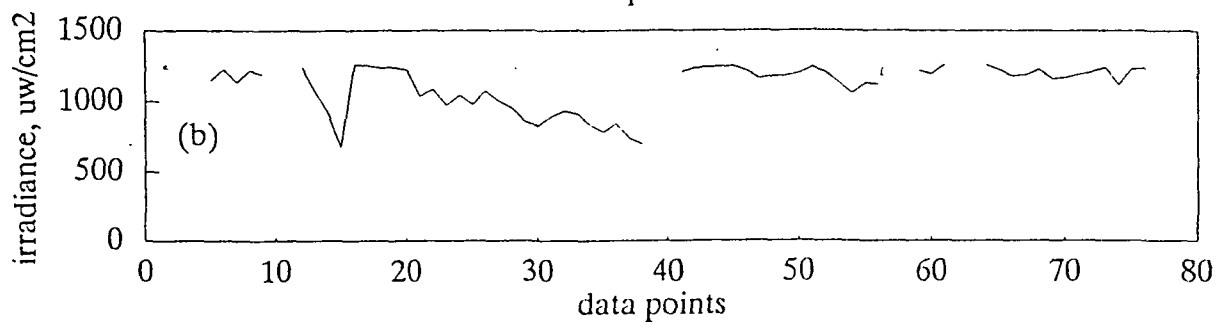


Fig. 6b

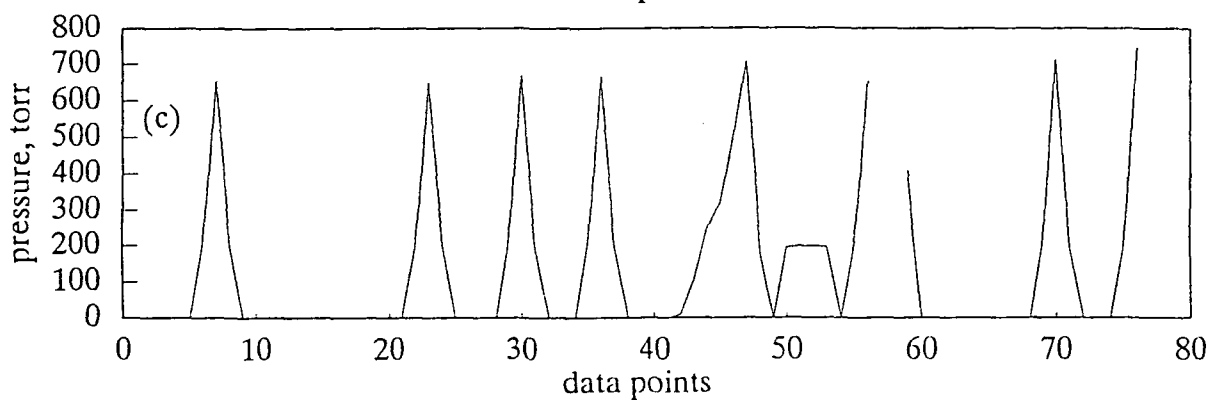


Fig. 6c

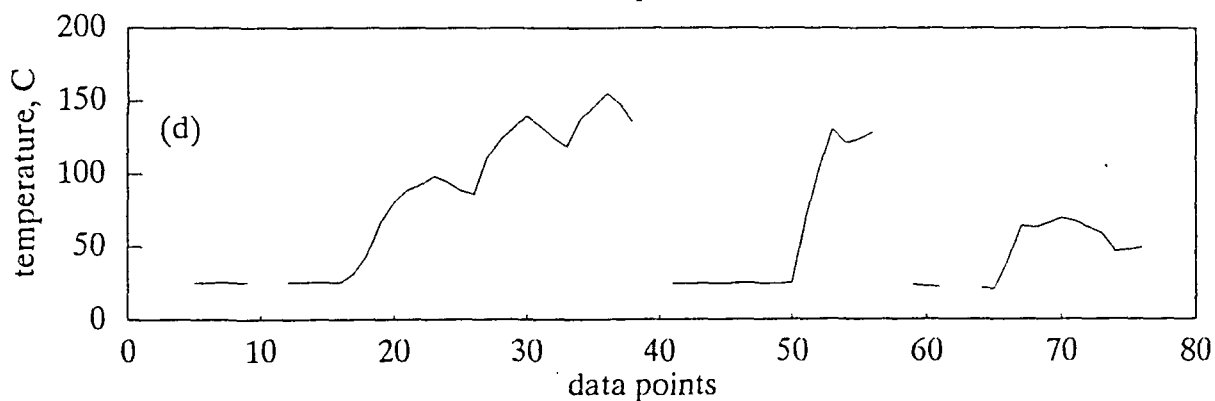


Fig. 6d

Fig. 6. - Experiment test conditions for luminescence measurements: (a) time, (b) UV monitor, (c) pressure gage and (d) thermocouple temperature.

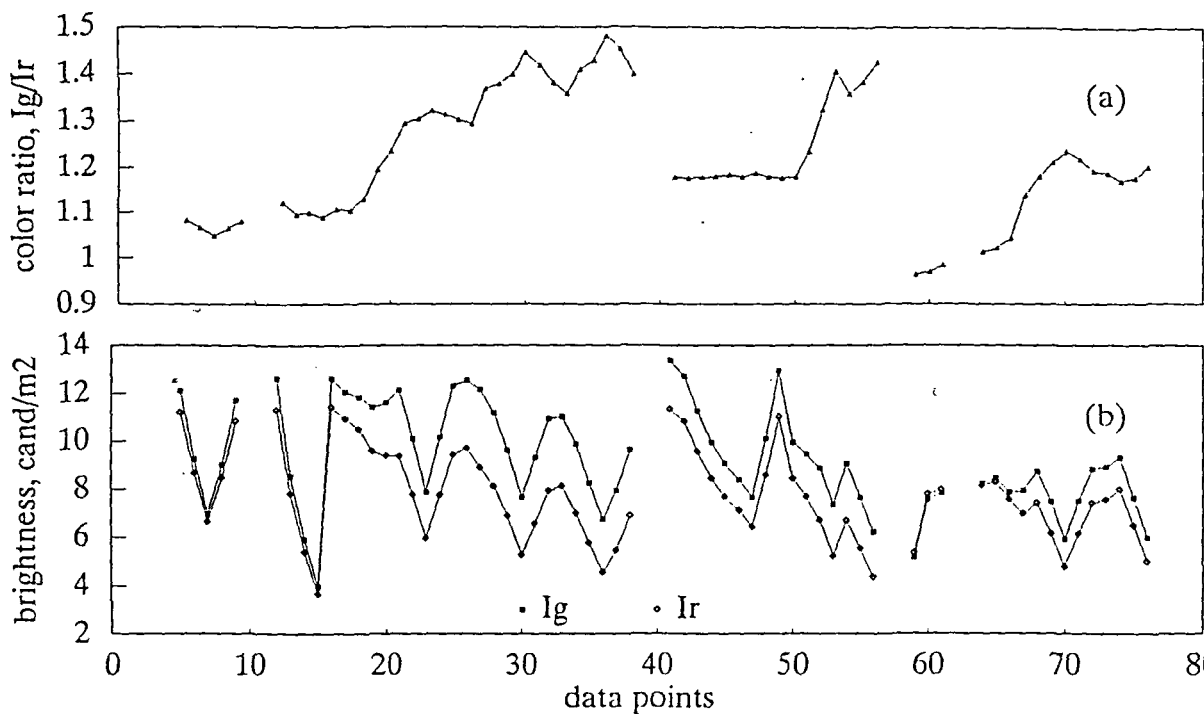


Fig. 7a

Fig. 7b

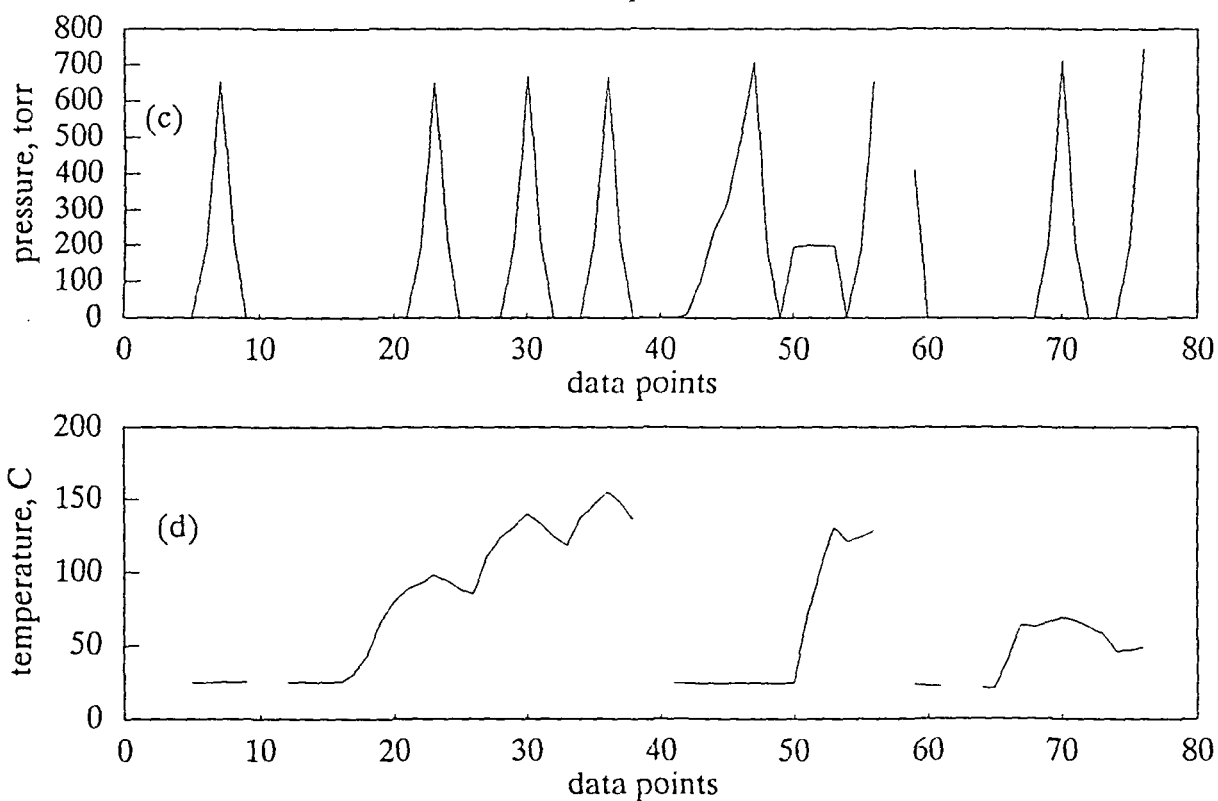


Fig. 7c

Fig. 7d

Fig. 7. - Calibrated outputs for sample region 2 (a) color ratio, (b) individual brightness, (c) pressure gage and (d) thermocouple temperature.

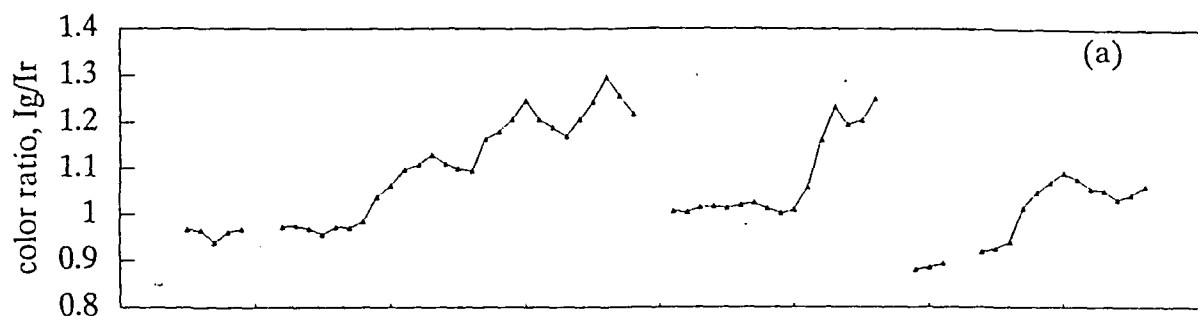


Fig. 8a

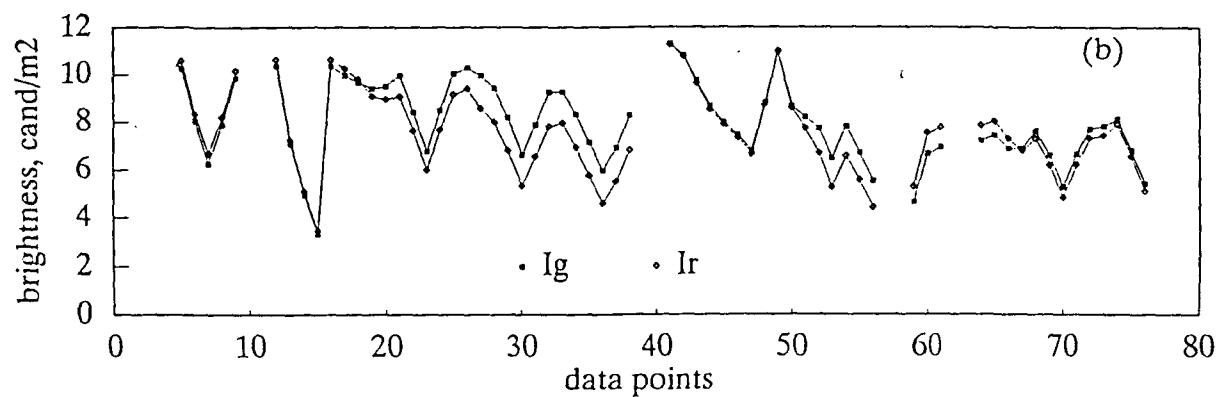


Fig. 8b

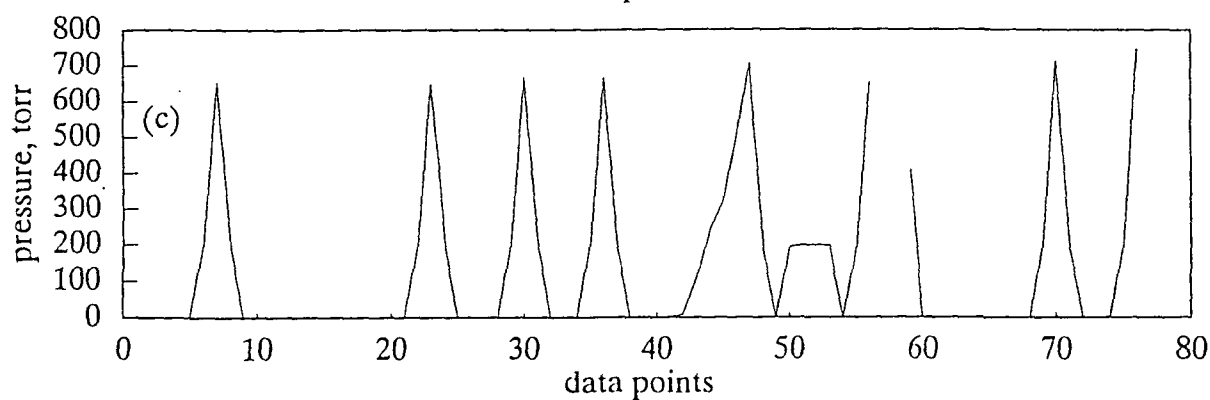


Fig. 8c

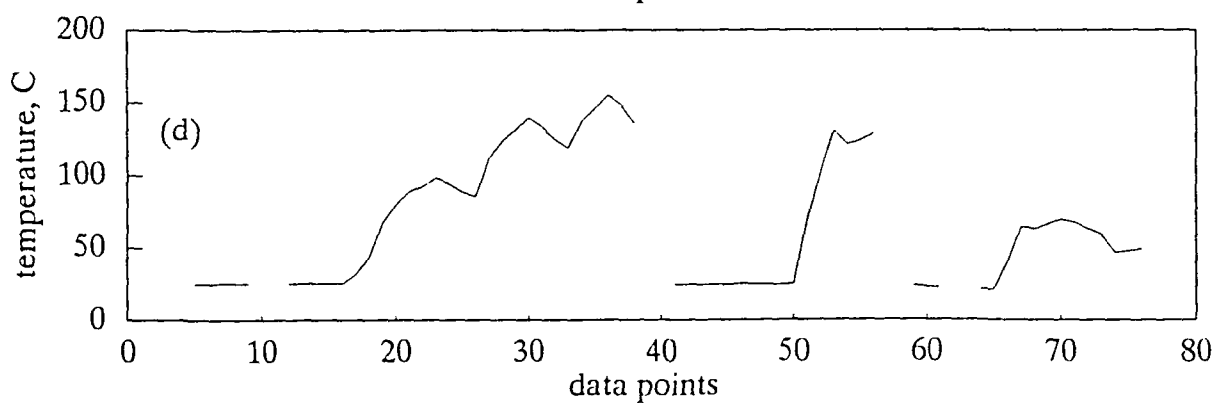


Fig. 8d

Fig. 8. - Calibrated outputs for sample region 3 (a) color ratio, (b) individual brightness, (c) pressure gage and (d) thermocouple temperature.

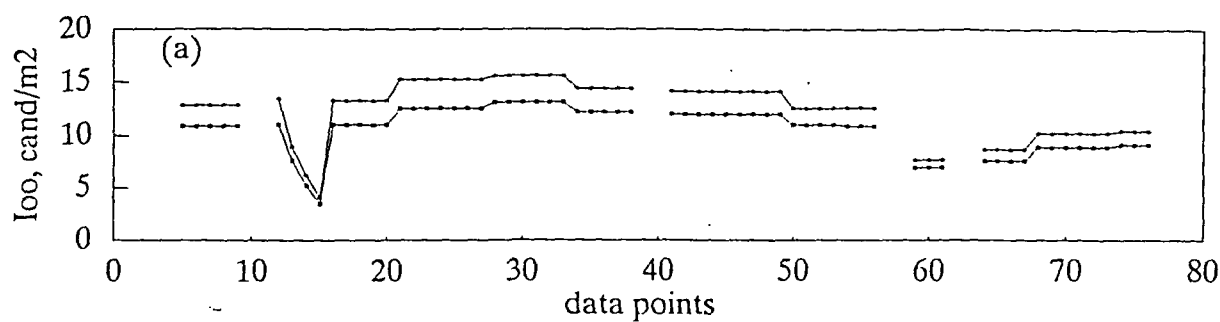


Fig. 9a

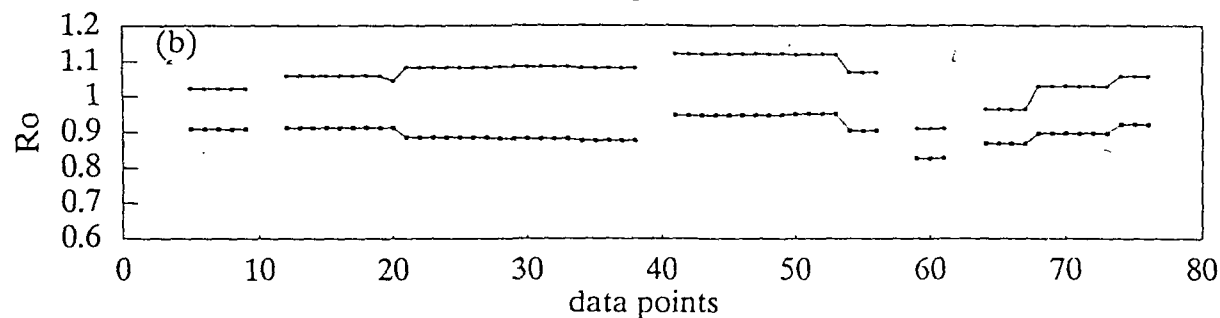


Fig. 9b

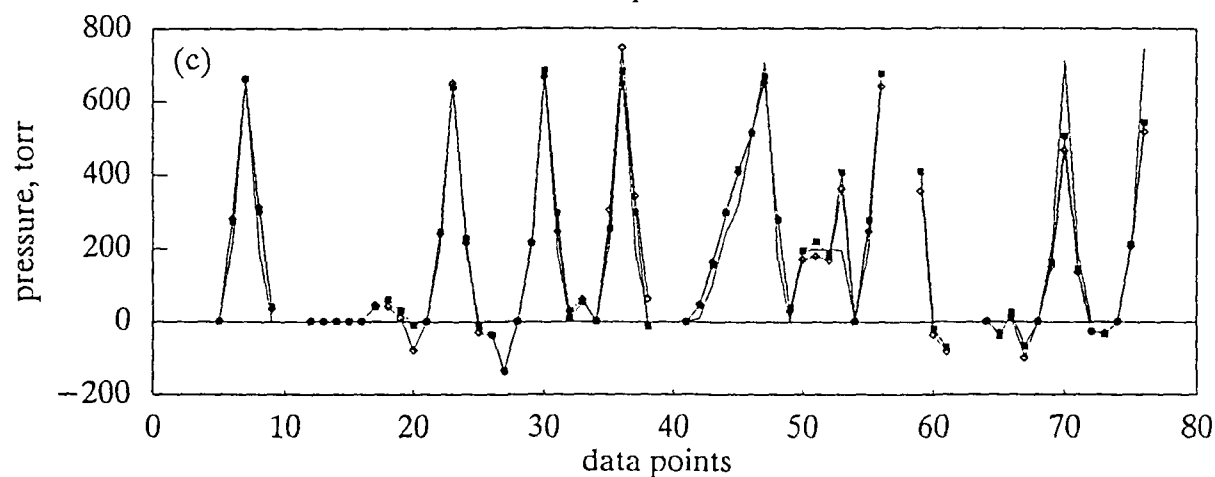


Fig. 9c

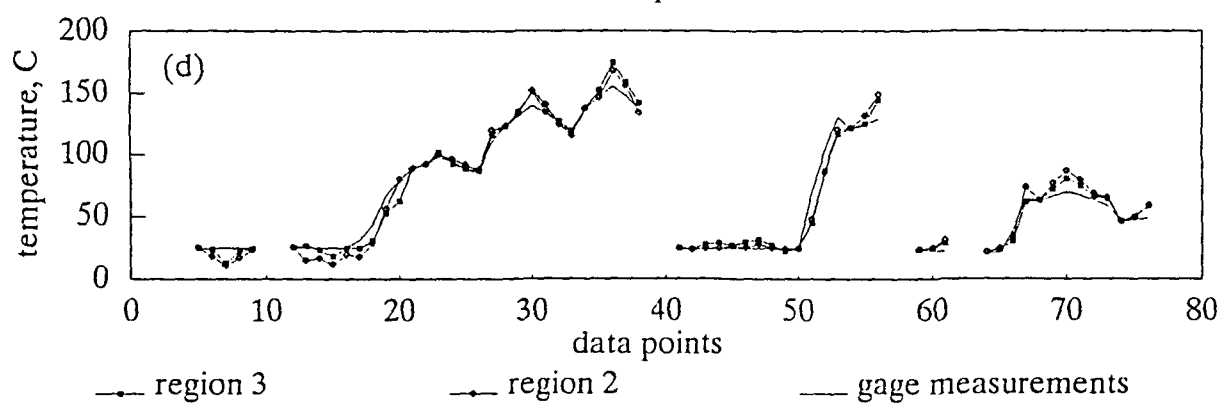


Fig. 9d

Fig. 9. - Processed values for luminescence pressure and temperature: (a) reference brightness, (b) reference color ratio, (c) luminescence pressure and (d) temperature values with instrumented gage comparisons.

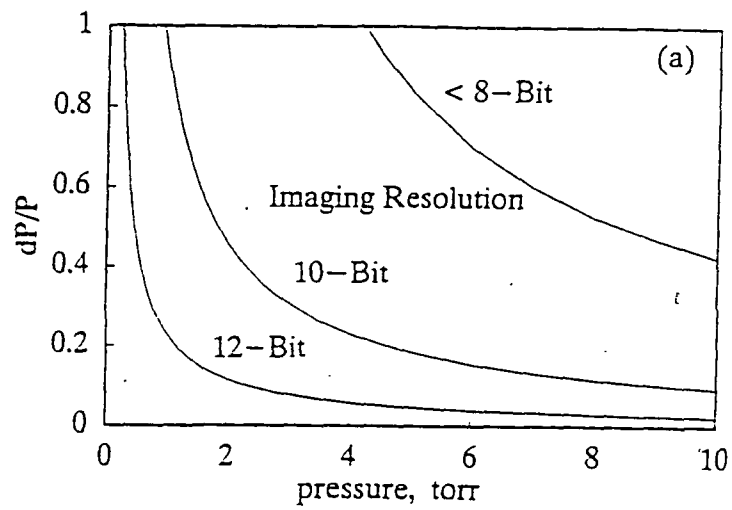


Fig. 10a

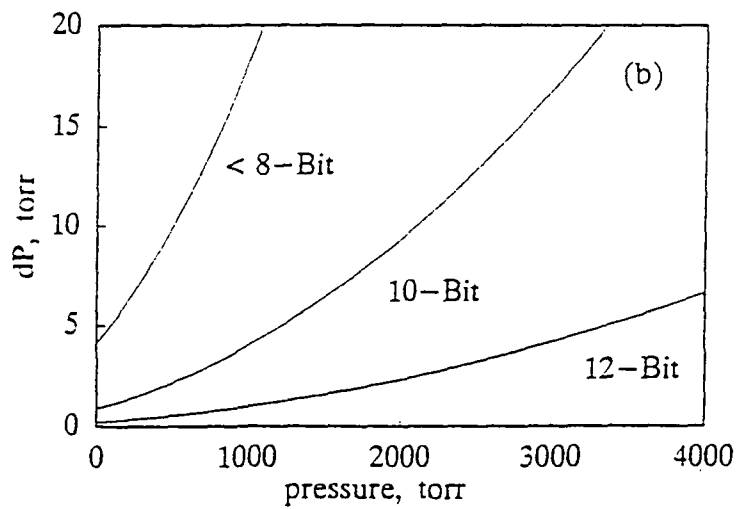


Fig. 10b